A B S T R A C T

A DAMPER FOR A LANDING GEAR LEG, AND LANDING GEAR HAVING INDEPENDENT LEGS FITTED WITH SUCH DAMPERS

The invention relates to a damper for an airplane landing gear leg, the damper being of the type comprising a main strut (11) and a rod-piston (13) co-operating with the strut (11) to define a main chamber (15) and annular chamber (16) for hydraulic fluid, and presenting internally two adjacent chambers (19, 20) that are isolated from each other by a separator piston (21). In accordance with the invention, the damper (10) further comprises a first secondary strut (26) telescopically slidable on the above-mentioned rod-piston (13), and a second secondary strut (37) telescopically slidable on the other end of the first secondary strut (26). The two second annular chambers (31, 40) as defined in this way are respectively connected to associated control circuits thus enabling the total length of the damper to be shortened or lengthened respectively for the purpose of causing the landing gear leg to contract or to be extended.

Translation of the title and the abstract as they were when originally filed by the Applicant. No account has been taken of any changes that may have been made subsequently by the PCT Authorities acting ex officio, e.g. under PCT Rules 37.2, 38.2, and/or 48.3.